

IN THE CLAIMS

✓ Please cancel claims 9-11.

Please replace the following claims with the claims of the same number:

1. An internal compression supersonic aircraft inlet comprising: an internal duct having an opening for receiving airflow and a throat section, said internal duct having one or more internal compression surfaces wherein substantially all of the compression of said airflow takes place within said inlet internal duct, said throat section of the inlet further incorporating a shock stability bleed system having one or more bleed passageways, wherein a portion of said airflow is removed from said internal duct through said one or more bleed passageways so that an airflow shock wave is maintained within said throat section.

2. An inlet according to claim 1 wherein said shock stability bleed system further comprises bleed passages having a variable area exit.

24 3. An inlet according to claim 1 wherein said throat section of said inlet further comprises movable sidewalls in the throat section for varying the throat area.

4. An inlet according to claim 1 wherein said internal duct has a rectangular cross-section.

5. An inlet according to claim 1 wherein the internal compression surfaces of said inlet duct are shaped to produce isentropic compression of the airflow.

6. An inlet according to claim 1, wherein said inlet further comprises exterior surfaces having a rectangular cross-section.

7. An inlet according to claim 1 wherein said inlet further comprises external surfaces that are aligned with the flow of air to the inlet.

8. An inlet according to claim 6 wherein said external surfaces are substantially parallel to the flow of air to the inlet.

Please add the following new claims:

12. The inlet of claim 1 wherein the interior surfaces of the internal duct further comprise a plurality of compression angled surfaces.

13. The inlet of claim 1 wherein said shock stability bleed system further comprises bleed passages having a fixed area exit.

14. An inlet according to claim 1 wherein said internal duct has an axi-symmetric cross-section.

15. An inlet according to claim 1 wherein said inlet further comprises exterior surfaces having a rectangular cross-section which transition to a round nacelle.

16. An inlet according to claim 1 wherein a portion of the interior surfaces of the internal duct have porous surfaces.

17. An inlet according to claim 1 wherein the interior surfaces of the internal duct have continuous surfaces from the opening to the exit of the inlet.

18. A supersonic inlet for use with an aircraft, said supersonic inlet comprising:

an internal duct having an opening for receiving airflow and a throat section, said internal duct having one or more internal compression surfaces wherein substantially all of the compression of said airflow takes place within said inlet duct,

said opening of said internal duct further comprised of a first and second leading edge, wherein said leading edges are staggered in location;

said throat section of the inlet further incorporating a shock stability bleed system, wherein a portion of said airflow is removed from said internal duct so that a shock wave is maintained within said throat section.

19. A supersonic inlet for use with an aircraft, said supersonic inlet comprising:

an internal duct having an opening for receiving airflow and a throat section, said internal